

IN THE CLAIMS:

1. (Currently Amended) A method of coating a gas turbine engine component using a powder coating process to produce a higher melting point coating comprising:
providing a gas turbine engine component having a solid surface;
applying a powder coating to the gas turbine engine component solid surface using the powder coating process, wherein the powder coating is applied in a dry form without an organic solvent, the powder coating comprising a fritted glass matrix with ceramic particles trapped in the matrix, and
heating the applied powder coating ~~in a single step~~ to only one temperature to raise the melting point of the glass of the matrix by reacting the ceramic particles with the glass thereby raising the melting point and resulting in a higher engine use temperature, wherein the heating to the only one temperature melts and fuses particles of the powder coating to the gas turbine engine component solid surface and cures the powder coating.
2. (Previously Presented) The method of claim 1, wherein the powder coating is applied using a fluidized bed process, an electrostatic spray process or an electrostatic brush process.
3. (Canceled)
4. (Previously Presented) The method of claim 2, wherein the gas turbine engine component is grounded.
5. (Original) The method of claim 4, wherein the powder coating comprises an inorganic based or organic based material.
6. (Previously Presented) The method of claim 5, wherein the powder coating comprises a material selected from the group consisting of a glass/enamel/metal and a composite.
7. (Previously Presented) The method of claim 6, wherein the powder coating comprises

a material selected from the group consisting of silica, alumina, zirconia, magnesium oxide, titanium oxide, yttrium and hafnium oxide.

8. (Original) The method of claim 5, wherein the coating is a thermal barrier coating.
9. (Original) The method of claim 5, wherein the gas turbine engine component is cleaned prior to application of the powder coating.
10. (Canceled)
11. (Previously Presented) The method of claim 1, wherein the component includes an electrically conductive non-metallic substrate.
12. (Currently Amended) A method of coating a gas turbine engine component using a powder coating process to produce a higher melting point coating consisting of ~~comprising~~:
providing a gas turbine engine component having an electrically conductive solid substrate;
cleaning the gas turbine engine component prior to application of a powder coating;
applying a powder coating to the solid substrate of the gas turbine engine component using the powder coating process, wherein the powder coating is applied in a dry form without an organic solvent, the powder coating comprising a fritted glass matrix with ceramic particles trapped in the matrix; and
heating ~~in a single step~~ the applied composition to only one temperature to raise the melting point of the glass of the matrix by reacting the ceramic particles with the glass thereby raising the melting point and resulting in a higher engine use temperature, wherein the heating to the only one temperature melts and fuses particles of the powder coating to the gas turbine engine component and cures the powder coating.
13. (Canceled)
14. (Original) The method of claim 12, wherein heat at a temperature between about 450-

1538°C for about 5 minutes to about 24 hours is applied.

15. (Canceled)

16. (Canceled)

17. (Previously Presented) The method of claim 1, wherein the powder coating is applied directly to the gas turbine engine component solid surface or the powder coating is applied to a bond coating located on the gas turbine engine component solid surface.

18. (Previously Presented) The method of claim 1, wherein a tribo charging process or a corona charging spray process is employed.

19. (New) The method of claim 1, wherein the component is coated and heated in less than 15 minutes.

20. (New) The method of claim 12, wherein the component is coated and heated in less than 15 minutes.

21. (New) A method of coating a gas turbine engine component using a powder coating process consisting of:

providing a gas turbine engine component having a solid surface;

applying a powder coating to the gas turbine engine component solid surface using the powder coating process, wherein the powder coating is applied in a dry form without an organic solvent, the powder coating comprising an organic material; and

heating the applied powder coating to only one temperature to melt and fuse particles of the powder coating to the gas turbine engine component solid surface and cure the powder coating.

22. (New) The method of claim 21, wherein the component is coated and heated in less than 15 minutes.

23. (New) A method of coating a gas turbine engine component using a powder coating process comprising:

providing a gas turbine engine component having a solid surface;

applying a powder coating to the gas turbine engine component solid surface using the powder coating process, wherein the powder coating is applied in a dry form without an organic solvent, the powder coating comprising a ceramic matrix admixed with metal; and

heating the applied powder coating to melt and fuse particles of the powder coating to the gas turbine engine component solid surface and cure the powder coating.

24. (New) The method of claim 23, wherein the applied powder coating is heated to only one temperature to melt and fuse particles of the powder coating to the gas turbine engine component solid surface and cure the powder coating.